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National Aeronautics and **Space Administration**

AEROTHERMAL MODELING

Phase I - Final Report **Volume II - Experimental Data**

> ρλ M.J. Kenworthy S.M. Correa D.L. Burrus

General Electric Company Aircraft Engine Group **Advanced Technology Operation**

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1.0 INTRODUCTION

The Aerotheral Modeling Program-Phase was conducted as part of the overall Hot Section Technology (HOST) Program initiated by NASA. The Phase I program was a 9 month effort involving the assembly of a benchmark quality data base from selected available literature, and from General Electric's engine and combustor component test data. Additional definitive data was obtained from an experimental test program conducted as part of the Phase I effort. Contained within this volume is a discussion of the experimental test effort, and a complete compilation of the test data obtained. The compilation is divided into sets representing each of the 18 experimental configurations tested. A detailed description of each configuration, and plots of the temperature difference ratio parameter or pattern factor parameter calculated from the test data are also provided.

2.0 DISCUSSION

As part of the Phase I Aerothermal Modeling Program, an experimental test program was conducted at General Electric. The purpose of this effort was to obtain additional definitive data with respect to the internal flow field of a combustor from which to assess the capabilities of the 3-D elliptic detailed internal flow model. A total of 18 configurations were tested using two available test rigs. Each test rig represented a 90° sector of a full annulus combustor.

The majority of the experimental testing was conducted in a rig representing a simple combustor design. This rig featured a flat uncooled dome inlet, and parallel cylindrical uncooled liner walls. The geometry of this rig was well within the geometrical constraints of the 3-D elliptic model. The dome section was interchangable to allow various dome configurations to be tested. In this testing effort, four dome inlet configurations were tested in this rig. These configurations were:

- Low pressure drop uniform/parallel flow inlet using a wire mesh
- High pressure drop uniform/parallel flow inlet using a perforated plate
- Nonswirl air injector cups
- Actual GE/F101 swirl cup hardware.

The liner walls featured two rows of dilution holes patterned after the GE/F101 combustor. The hole sizes could be varied or entirely blocked off depending on the desired configuration. Since this rig was totally uncooled, only two-temperature trace experimental testing was conducted using this rig.

The parallel wall test rig was designed with three separate flow control systems to independently set conditions feeding the dome inlet, the outer dilution holes, and the inner dilution holes. This provided the capability for two-temperature trace testing.

Data were obtained by using a rake with 21 equally (radial) spaced thermocouple elements mounted onto a traversing arm designed to allow axial as well as circumferential movement of the rake. During a typical test run, the rake was moved from -18° from top center to +18° from top center at 1° increments at 4 axial planes downstream of the aftmost dilution injection point. This provided an array of 777 temperature measurements within the 36° section at each axial plane. In addition to the thermocouples, 11 equally (radial) spaced impact pressure probes were initially mounted onto the rake. These pressures were used to map out the inlet velocity profiles in the rig when using the uniform/parallel flow dome inlet configurations. All data obtained were sent through a data logger system and loaded onto a cassett tape for posttest processing.

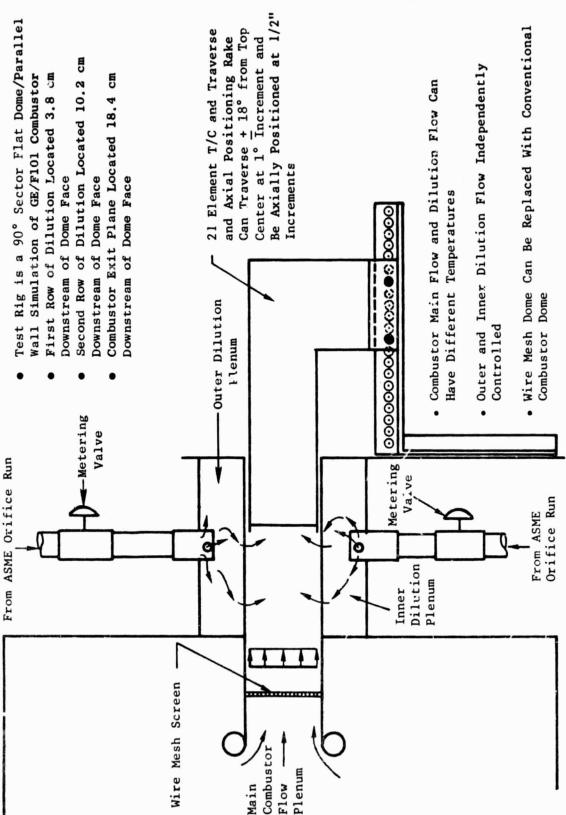
A schematic of the parallel wall test rig is shown in Figure 1. An overall view of the test rig hardware and closeups views of various features of the rig are shown in the photographs in Figures 2 through 5.

The other test rig used in the experimental test effort was a 90° sector of an actual GE/F101 combustor. The F101 combustor features a conical dome with 20 equally spaced counterrotating swirl cup assemblies. The liners are a machined-ring type featuring 6 film cooling slots in the outer liner, and 5 film cooling slots in the inner liner. The liners are contoured resulting in a converging interne' flowpath. Each liner has 2 rows of dilution holes. Corresponding rows on each liner feature opposed holes of equal size. The primary injection row features 40 holes equally spaced around the circumference, one hole directly in line with the cup center, and one hole between cup centers. The holes in line have a diameter of 0.84 cm (0.33 inch), while the holes between are larger with a diameter 1.04 cm (0.41 inch). The secondary injection row features 60 holes equally spaced around the circumference. The holes are offset 3° from the cup centers. All 60 holes have a diameter of 0.91 cm (0.36 inch).

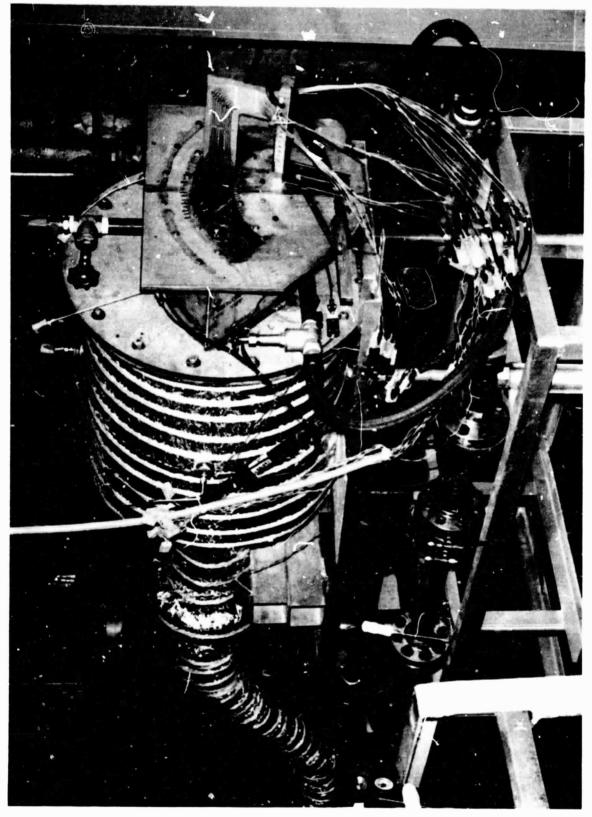
With this test rig, data were obtained by using a rake with 13 equally (radial) spaced thermocouple elements. The rake was mounted onto a traversing arm that permitted movement only in the circumferential sense. Thus temperature measurements in this rig were made only at the discharge plane. During a typical test run, the rake was moved from -36° from top center to +36° from top center in 1.5° increments. This provided an array of 637 temperature measurements within the 72° section of the discharge plane. All data obtained were sent through a data logger system and loaded onto cassett tapes for post-test processing. An illustration of the F101 sector combustor test rig is shown in Figure 6.

Of the 18 experimental configurations tested, 14 were conducted as two-temperature trace experiments in the parallel-wall test rig. The remaining 4 experimental configurations were tested in the F101 sector combustor test rig. Of these, two were conducted as two-temperature trace experiments and two were conducted with fuel and heat release. This experimental test series started with the simple case of a single row of uniform jets penetrating into a uniform/parallel crossflow, and progressed in complexity to cases in actual combustor hardware with fuel injection and heat release. This approach permitted investigating the effects of such phenomena as opposing rows of dilution jets, the introduction of swirl at the dome inlet, actual flowpath contours, and the introduction of fuel and heat release. A summary of the experimental configurations is provided in Table I. The test data obtained from this program provided an excellent source from which to make comparisons with calculations of the experimental configurations performed on the 3-D elliptic detailed internal flow model.

All of the test data obtained are contained along with a detailed description of each experimental configuration and the test conditions in Section 3.0. Also included in this section are contour plots of the temperature difference ratio for all two-temperature trace experimental test runs, and contour plots of the pattern factor parameter for all experimental test runs



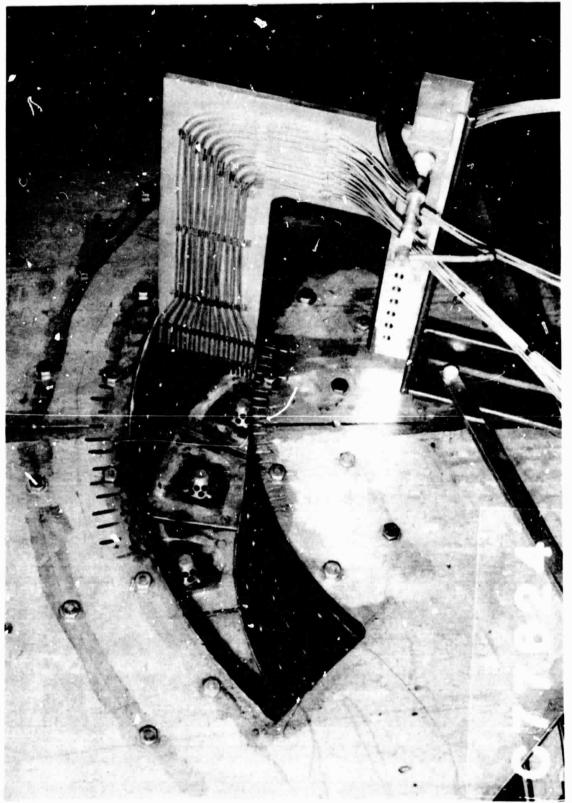
Aerothermal Modeling Experimental Test Rig (Unfueled). Figure 1.



Aerothermal Modeling Experimental Test Rig (Unfueled). Figure 2.



Aerothermal Modeling Experimental Test Rig (Unfueled). Figure 3.



Aerothermal Modeling Experimental Test Rig (Unfueled). Figure 4.



Aerothermal Modeling Experimental Test Rig (Unfueled). Figure 5.

ORIGINAL PAGE IS OF POOR QUALITY Outer Liner-Exhaust Gas Exit Dome-Rake Air Flow. Fuel. Inner Liner Tube Swirl Cup.

Figure 6. Lerothermal Modeling Experimental Test Rig (Fueled).

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Injection 233 M/Sec 5% Pressure Dr.p. SI Pressure Drop 51 Pressure Drop Drop entum Ratto 160 58 67 1 Pressure 53 3 3 3 3 3 3 3 2 270 Injection 2 Injection Injection Injection In sect 10n Injection JP-5 Low aP I JP-5 High &P Injectors f/a = 0.016 JP-5 High AP Injectors f/a = 0.012 JP-5 high aP Injectors f/s = 0.0.0 JP-5 Low ap Injectors f/s = 0.020 City Ges Low AP Injectors f/a = 0.012 Fuel Type 100 Cooling Cylindrical - Uncoled Standard Configurat Standard Configurat Well Type non Standard Configuration in ject or a Injectors Cooling Standard Configurat Perforated Plate SI AP Counterrotating Swirl Cup Dome Type AP Screen Sonswir! Air Standard Configuration (Equivalent Areas) 0.92 CM Dissector Specing Every 6 From 3 CM Specing avery 6 From 3 CM 0.513 CM Diameter Specing Every 6" From 3" CM 5.913 CM Dissertor Specing Every 6 From 3 CM From 3° CW Inner Well 2nd Dilurion Row Spacing Every 6" From 3' CW 3 3 0.794 CM Diameter Spacing Every 6" From 3" CM From 3" CW From 3" CM Spacing Every 6" From 3" Standard Configuration (Equivalent Areas) 0.93 CM Dismeter Specing Every 6 From 3 Spacing Every 6 From 3 Outer Well Ind Dilution Row Spacing Every 6" Spacing Every 6 . 20. .99 . 95 2.5 18. Standard Configuration (Equivalent Areas) 0.794 CM Diameter 0', 18 , 1.032 CM Diameter 9', 27', 18. 27. Top Inner Well lat Dilution Row 0,794 CM Diameter Spacing Every 9 From 1.032 CM Diameter 9", CM Diameter 0'. 0.794 CM Diameter 0. 1.032 Standard Configuration (Equivalent Areas) U 794 CM Dismete: 0°, 18°, 36° 1.032 CM Dismeter 9°, 27°, 45° .99 1.6 Top Outer Wa'l CM Diameter 9. 0.794 CM Diameter Spacing Every 9" F 0.794 (15 (GE/F101 Sector) 17A (GE/F:01 Sector) 178 (GE/FI01 Sector) 13B (GK/F101 Sector) GE/FIGI Sector) 17D (GE/F161 Sector) 18A (GE/F101 Sector) 168 (GE/F101 Sector) 13C (GE/F101 Sector Configuration 18D (GR/F101 Sect 0.7

10

Summary of Experimental Test Configurations.

Table I.

with fuel and heat release. The definitions of these nondimensional temperature parameters are provided below.

Temperature Difference Ratio:

$$\theta = \frac{T_{Inlet} - T_{Local}}{T_{Inlet} - T_{Jet}}$$

Pattern Factor

$$P_{f} = \frac{T_{Local} - T_{Inlet}}{T_{Avg} - T_{Inlet}}$$

For confiugrations involving the GE/F101 sector combustor, Table II provides the measured flow areas of each flow feature of the GE/F101 combustor. The areas in this table are representative of a full annular combustor.

Table II. F101 Combustor Airflow Feature Flow Characterisitics (Full Annular).

	Flow Area	Area Distribution %
Outer Liner		
Primary Dilution	3.403	7.87
In Line	1.361	3.15
Between	2.041	4.72
Secondary Dilution	3.932	9.10
Inner Liner		
Primary Dilution	3.403	7.87
In Line	1.361	3.15
Between	2.041	4.72
Secondary Dilution	3.932	9.10
Swirl Cup	6.800	15.73

3.0 EXPERIMENTAL TEST DATA COMPILATION

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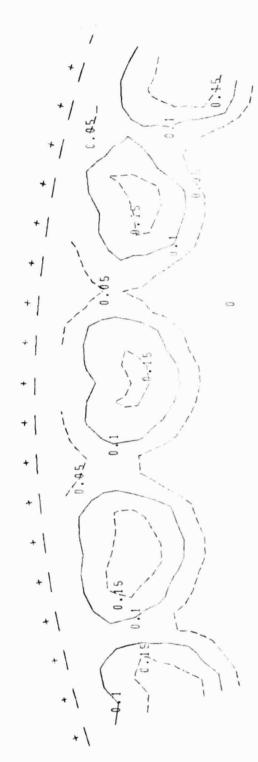
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Configuration 1, Single Row of Outer Jets into Uni Crossflow 2.75 inches Downstream of Primary Injection Point

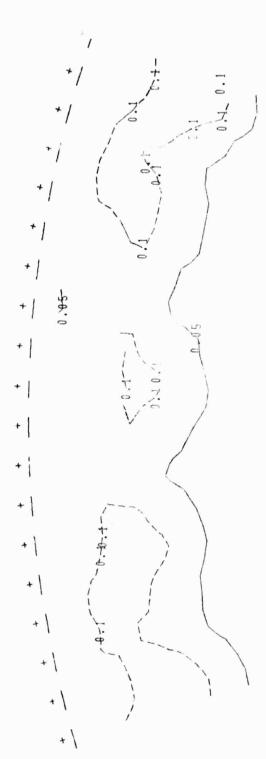


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Configuration 1, Single Row of Outer Jets into Uni Crossflow 5.75 inches Downstream of Primary Injection Point



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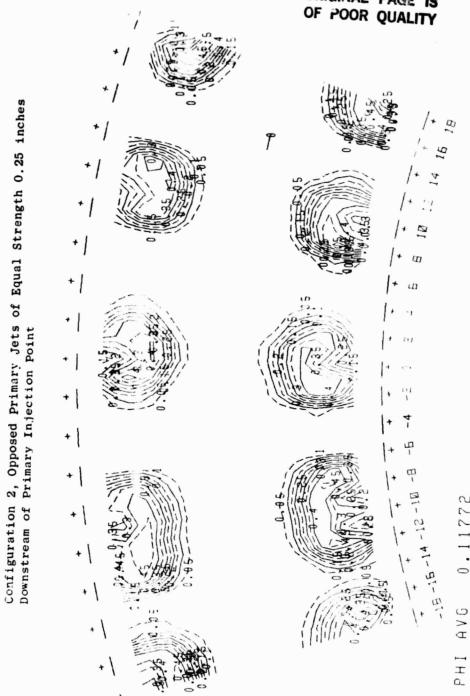
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ATTERN: P WALL: SPACING EVERY 9 HCLES 5/16 INCH DIAMETER WALL: SPACING EVERY 9 D DLES 5/16 INCH DIAMETER CMOMENTUM RATIO - 1 JET FLOW: 0.10 PPS AT (MOMENTUM
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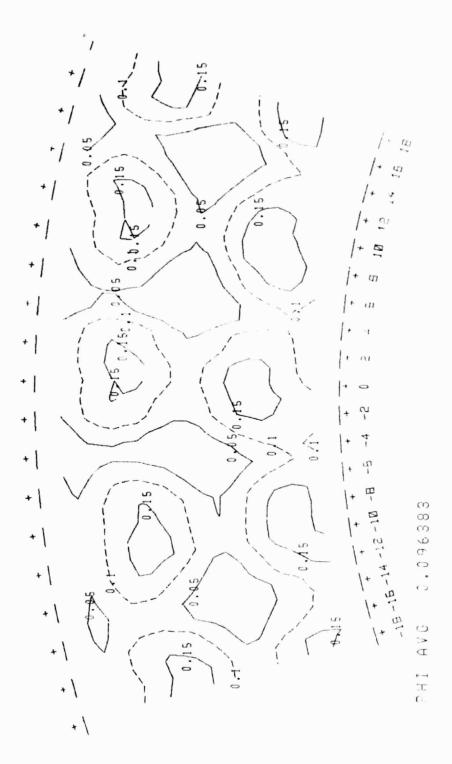
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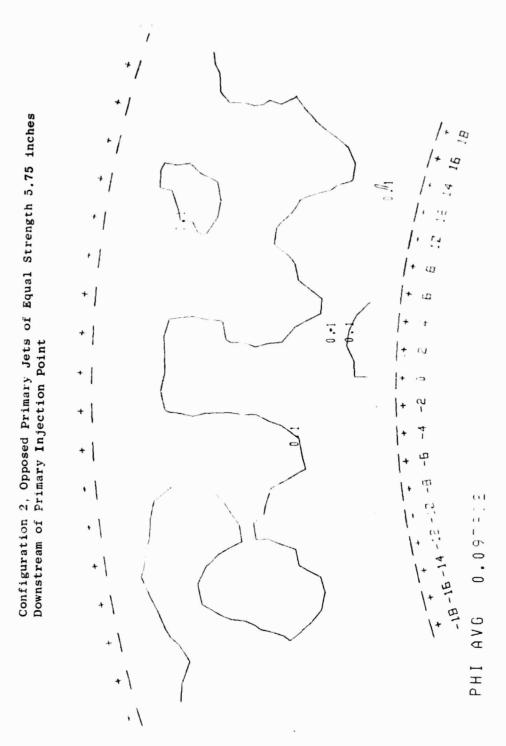
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Configuration 2, Opposed Primary Jets of Equal Strength 2.75 inches Downstream of Primary Injection Point





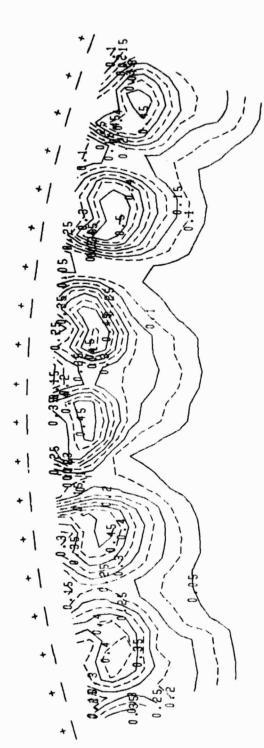
EXPERIMENTAL CONFIGURATION 3 ALL JET ROWS INTO A UNIFORM	DUTER WALL: PRIMARY HOLES SPACED EVERY & DEG STARTING AT TOP CENTER; ALL HOLES 5/16 INCH DIAMETER SECOND/RY HOLES SPACED EVERY 6 DEG STARTING AT 3 DEG CW FROM TOP CENTER; ALL, HOLES 5/16 INCH DIAMETER BY SECOND CONTER; ALL, HOLES 5/16 INCH DIAMETER	PAGE IS QUALITY	0 DEG R, VELOCITY = 128 FPS T 531 DEG R, VELOCITY = 353 FPS	15:1)	MEASURED GAS TEMPERATURES	DEGREES R	SECONDARY INJECTION POINT	0K4 0K4 0K2 04K 026 000 007 006 007 070 000	955. 954. 951. 943. 929. 913. 900. 888. 885. 877. 872. 969. 869.	955 954 950 941 926 909 897 888 875 836 788 773 774 810	957. 957. 956. 954. 951. 946. 938. 930. 917. 991. 881. 843. 805. 797. 784. 801. 794.	956 956 955 952 948 939 971 919 910 899 893 884 879 879 885	956, 956, 954, 948, 941, 927, 913, 898, 884, 863, 847, 824, 816, 816, 847, 956, 956, 954, 948, 939, 924, 909, 891, 872, 843, 811, 776, 766, 770, 805,	925, 932, 947, 939, 926, 911, 886, 860, 628, 790, 757, 760, 783, 831, 08/22/83 8.144 PAGE 1 ***	
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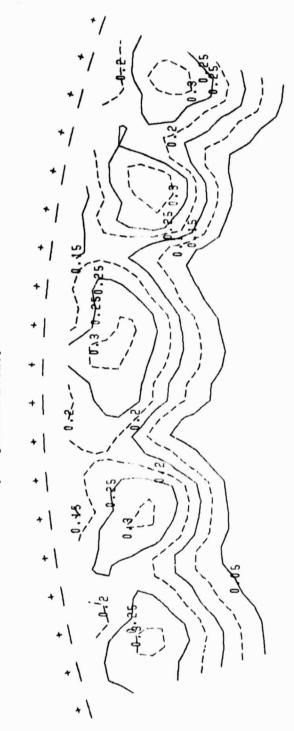
Configuration 3, Staged Outer Liner Dilution of Equal Strength 0.25 inches Downstream of Secondary Injection Point.



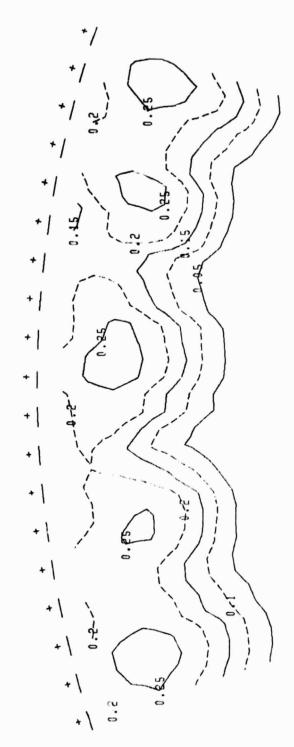
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Configuration 3, Staged Outer Liner Dilution of Equal Strength 1.25 inches Downstream of Secondary Injection Point.

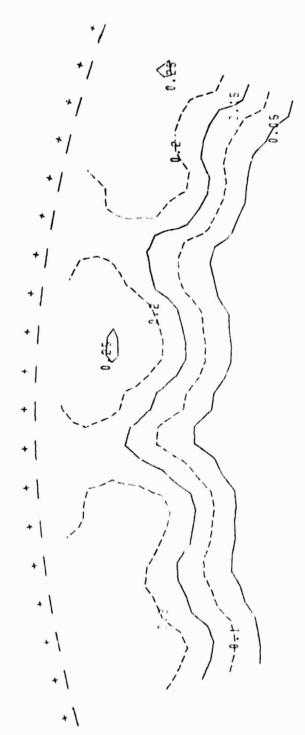


Configuration 3, Staged Outer Liner Dilution of Equal Strenght 2.25 inches Downstream of Secondary Injection Point.



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Configuration 3, Staged Outer Liner Dilution of Equal Strenght 3.25 inches Downstream of Eecondary Injection Point.



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HOLE PATTERN: OPPOSED PRIMARY JET ROWS INTO A UNIFORM HOLE PATTERN:

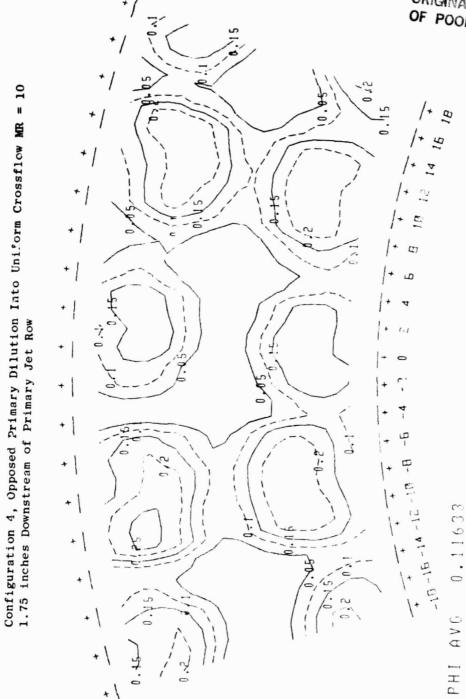
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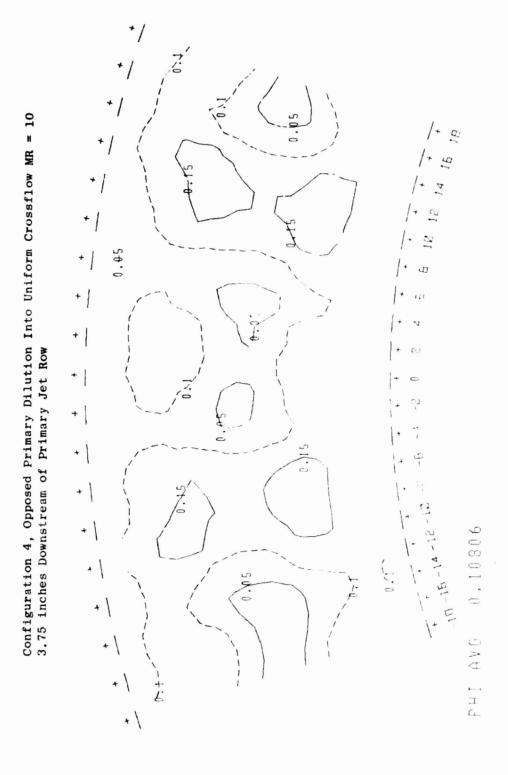
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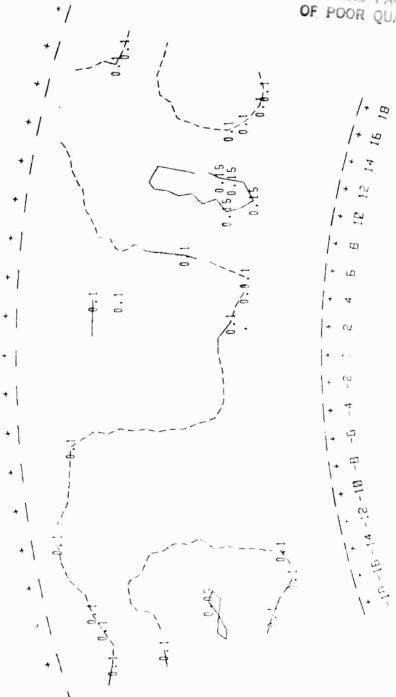
Configuration 4, Opposed Primary Dilution Into Uniform Crossflow MR = 10 0.25 inches Downstream of Primary Jet Row 18 D + | -18-16-14-12-18-8 0.1246 AVG PH1 * \

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Configuration 4, Opposed Primary Dilution Into Uniform Crossflow MR = 10 5.75 inches Downstream of Primary Jet Row

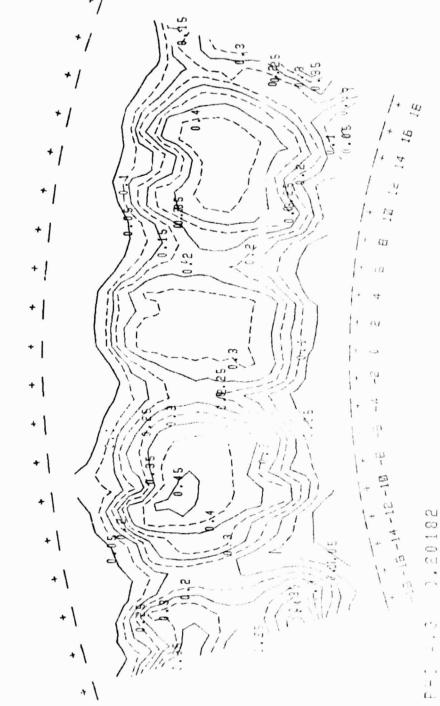
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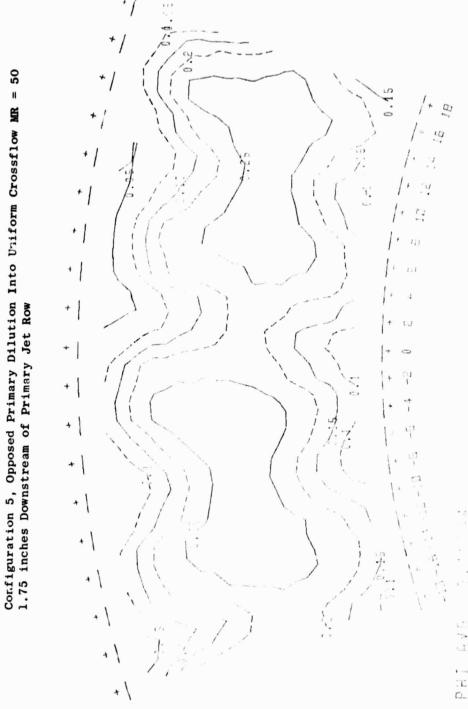
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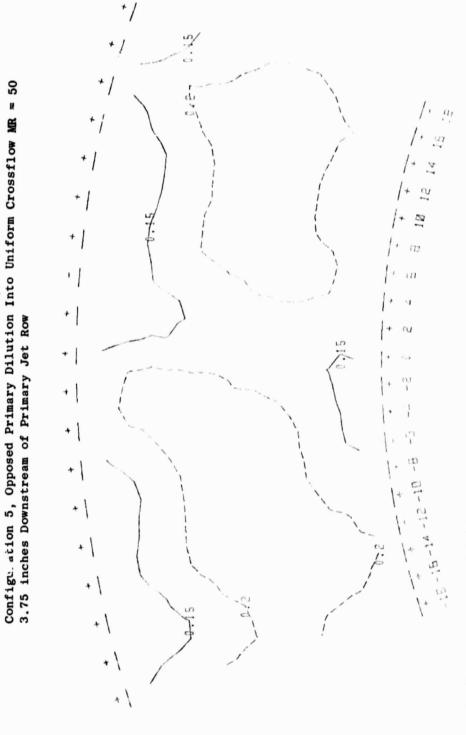


Configuration 5, Opposed Primary Dilution Into Uniform Crossflow MR = 50

0.25 inches Downstream of Primary Jet Row

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Configuration 5, Opposed Primary Dilution Into Uniform Crossflow MR = 50 5.75 inches Downstream of Primary Jet Row

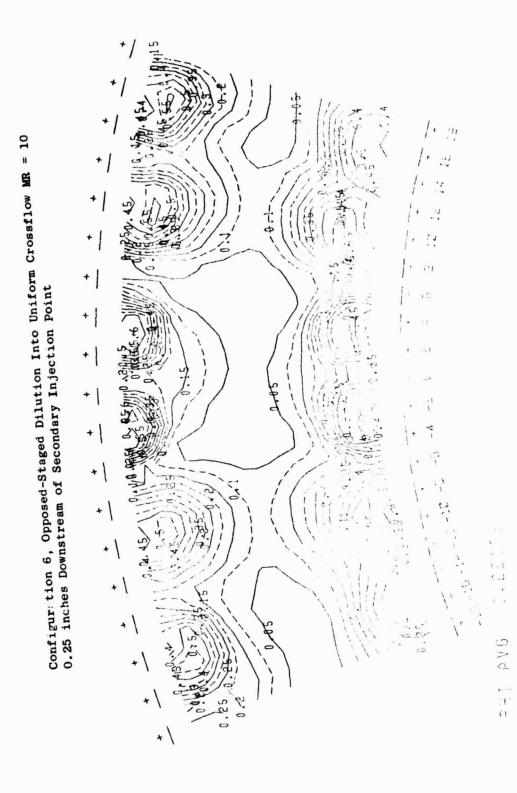


EXPERIHEN	
DESCRIPTION: OPPOSED AND STAGED JET ROWS INTO A UNIFORM //PARA!EL_CROSSFLOW	
HOLE PATTERN: GUTER WALL: PRIMARY HOLE SPACING EVERY 9 DEG STARTING AT TOP CENTER;	
HOLES AT 0, 18, 36 DEG 5/16 INCH DIAMETER HOLES AT 9,27,45 DEG 13/32 INCH DIAMETER SECRETARY HOLE SPACING EVERY 6 DEG STARTING 3 DEG CW FROM TABLORDARY HOLE SPACING ALL HOLES 23/64 INCH DIAMETER	
PRIMAPY HOLE SPACING EVERY 9 18,36 DEG 5/16 INCH DIAM 27,45 DEG 13/32 INCH DIA	т
OF	
CROSSFLOW: 2.07 PPS AT 960 DEG R, VELOCITY = 127 FPS OUTER JET FLOW: 0.26 PPS AT 543 DEG R, VELOCITY = 293 FPS	
(MOMENTUM RATIG - 10:1)	
INNER JET FLOW: 0.26 PPS AT 539 DEG R, VELOCITY = 293 FPS	_
MEASURED GAS TEMPERATURES	
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0.25 INCHES DOWNSTREAM OF SECONDARY INJECTION POINT	
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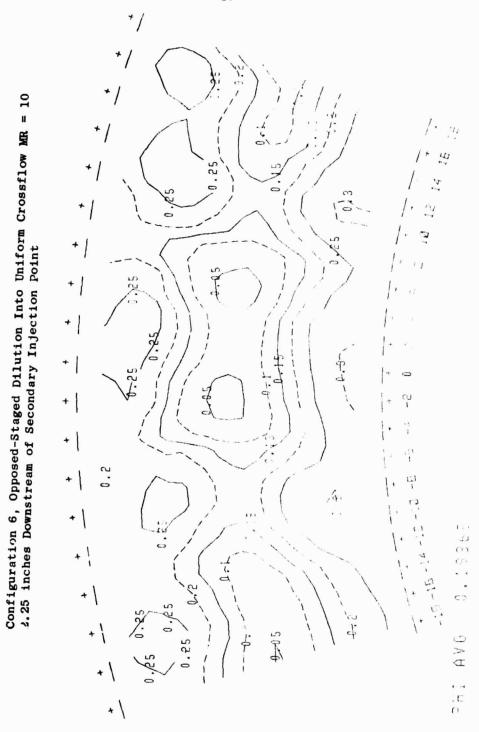
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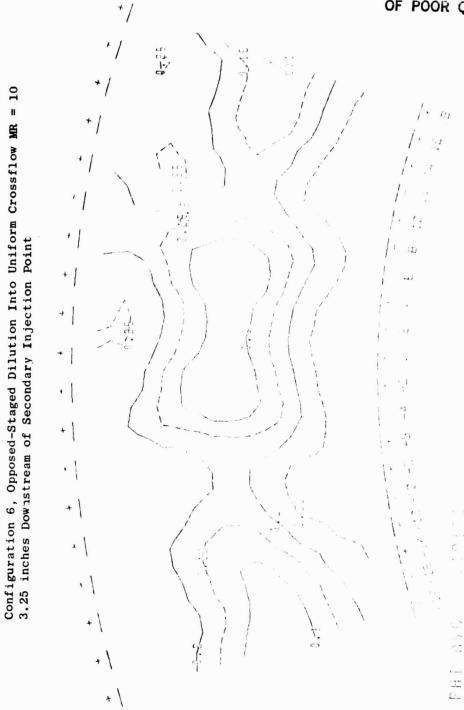
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Configuration 6, Opposed-Staged Dilution Into Uniform Crossflow MR = 10 1.25 inches Downstream of Secondary Injection Point HΛ Д Ц



ORIGINAL PACE IS OF POOR QUALITY

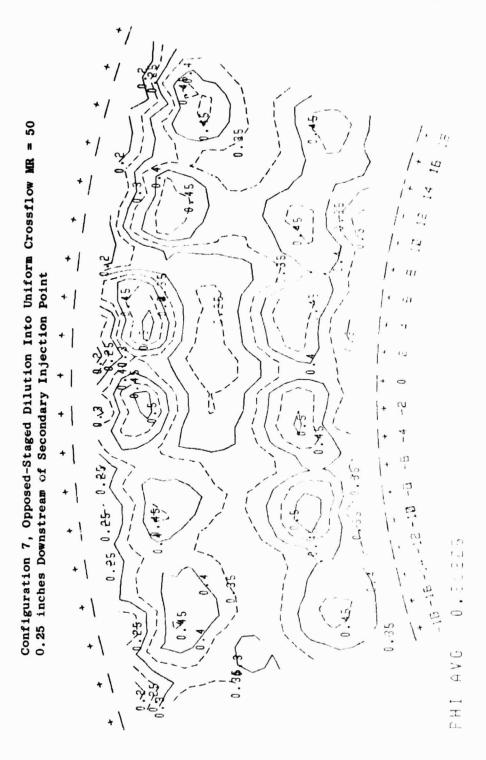


PARAMETER CROSSING OFFICE ON STARED JET RONS INTO A INIFCRM HOLE PATTERS: HOLE PATTERS: HOLE PATTERS: HOLE PATTERS: HOLE STAND HOLE STAND HOLE STAND HOLE STAND AT TOP CENTER; HOLE STAND 18-38 DEC	EXPERIMENTAL CONFIGURATION 7
The CRITTER FRIMARY HOLE SPACING EVERY 9 DEG STRATING AT TOP CENTER;	OPPOSED AND STAGED JET ROWS INTO DSSFLOW
SECONDARY HOLE SPACING EVERY 6 DEG STAFFTING OFFICE	PRIMARY HOLE SPACING EVERY 9 DEG STARTING
HINER WILL: PRINCHE PROTONE FUENCY DEG STAFTING AT TOP CENTER; HOLES AT O. 18-36 DEG 5/16 INCH CIAMETER HOLES AT O. 18-36 DEG 5/16 INCH CIAMETER HOLES AT O. 18-36 DEG 5/16 INCH CIAMETER TOP CENTER; ALL HOLES SACCING EVERY 6 DEG STAFTING 2 DEG CV FP.H TOP CENTER; ALL HOLES SACCING EVERY 6 DEG STAFTING 2 DEG CV FP.H TOP CENTER; ALL HOLES SACCING EVERY 6 DEG STAFTING 2 DEG CV FP.H HER JET FLOW: 0.35 PPS AT 959 DEG R, VELOCITY = 292 FPS HER JET FLOW: 0.25 PPS AT 954 DEG R, PS AT 954 DEG R, 955 DE	- 13/32 INCH DIAMETER ERY 6 DEG STARTING 3 DEG CU 764 INCH DIAMETER
TOWNSHIP HOLES SOLING EVERY 6 DEP STARTING 3 DEG CH FR M TOWNSHIP HOLES SOLING EVERY 6 DEP STARTING 3 DEG CH FR M TOWNSHIP HOLES SOLING EVERY 6 DEP STARTING 3 DEG R, VELOCITY = 58 FPS (HOMENTUM RATIO - 48:1) HERAURED GAS TEMPERATURES (HOMENTUM RATIO - 48:1) HERAURED GAS TEM	TING AT TOP
ACTION Control Contr	LE SPACING EVERY 6 DE© STARTING 2 DEG CW ALL HOLES 23/64 INC'' DIAMETER
NEW JET FLOW: 0.25 PPS AT 534 DEG R, VELOCITY = 292 FPS NEW FLOW: 0.25 PPS AT 534 DEG R, VELOCITY = 292 FPS NEW FLOW: 0.25 PPS AT 534 DEG R, VELOCITY = 292 FPS NEW FLOW: 0.25 PPS AT 534 DEG R, VELOCITY = 292 FPS NEW FLOW: 0.25 PPS AT 534 DEG R, VELOCITY = 292 FPS NEW FLOW: 0.25 PPS AT 534 DEG R, VELOCITY = 292 FPS NEW FLOW: 0.25 PPS AT 545 PPS AT	3 PPS AT 959 DEG R, VELOCITY = 58 FPS
O 25 INCHES DOWNSTREAM OF SECONDARY INJECTION POINT 1 2 3 4 5 6 7 8 9 10 11 12 13 14 15 16 17 18 19 20 815. 814. 813. 814. 813. 814. 814. 815. 814. 815. 817. 819. 821. 827. 839. 840. 844. 849. 819. 815. 817. 819. 821. 827. 839. 840. 844. 839. 819. 815. 817. 819. 821. 827. 839. 840. 844. 839. 819. 815. 817. 819. 821. 827. 839. 840. 844. 839. 819. 812. 813. 819. 812. 813. 819. 812. 813. 819. 812. 813. 819. 812. 813. 819. 812. 813. 819. 821. 827. 839. 840. 844. 839. 849. 849. 849. 849. 849. 849. 849. 84	0.25 FPS AI 534 DEG K, VELOCIIT = 252 FPS M RATIO - 48 :1)
HEASURED GAS TEMPERATURES DEGREES R DEGREE	JET FLOW: 0.25 PPS AT 534 DEG R. VELOCITY = 292 FPS
DEGREES RECONDARY INJECTION POINT 1 2 3 4 5 6 7 8 9 10 11 12 13 14 15 16 17 18 19 20 812. 814. 813. 813. 812. 812. 812. 812. 813. 813. 815. 817. 819. 822. 827. 830. 838. 841. 846. 815. 817. 818. 819. 822. 827. 830. 838. 841. 846. 843. 842. 842. 843. 842. 843. 843. 843. 843. 843. 843. 843. 843	(MOMENTUM RATIO - 48 :1)
0.25 INCHES DOWNSTREAM OF SECONDARY INJECTION POINT 1 2 3 4 5 6 7 8 9 10 11 12 13 14 15 16 17 18 19 20 812. 814. 813. 812. 812. 812. 812. 813. 813. 815. 817. 819. 822. 827. 830. 839. 841 846. 815. 817. 818. 819. 821 827. 830. 830. 830. 841 842 842 842 843 842 842 842 843 843 844 842 842 843 815. 815. 815. 815. 814. 815. 814. 816. 815. 814. 816. 817. 818. 821 825. 823 833 840. 843 842 843 843 843 843 843 843 843 843 843 843	
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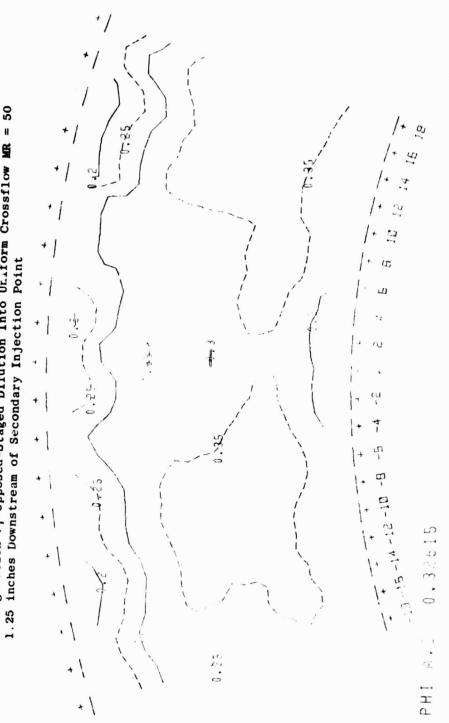
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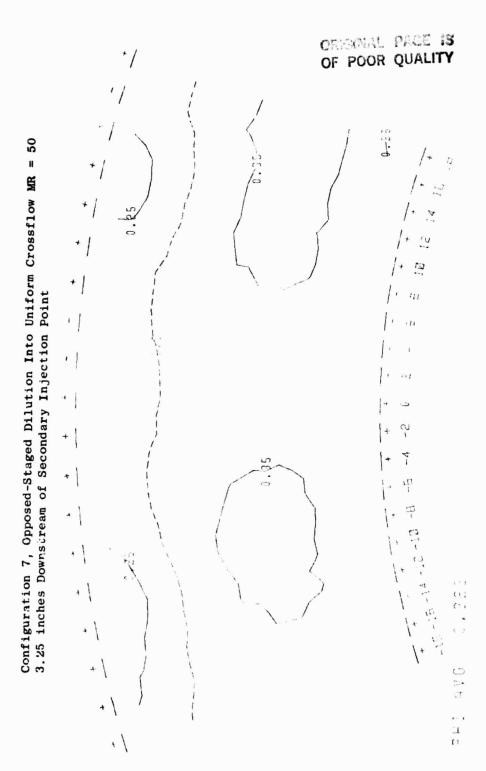
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Configuration 7, Opposed-Staged Dilution Into Uniform Crossflow MR = 50

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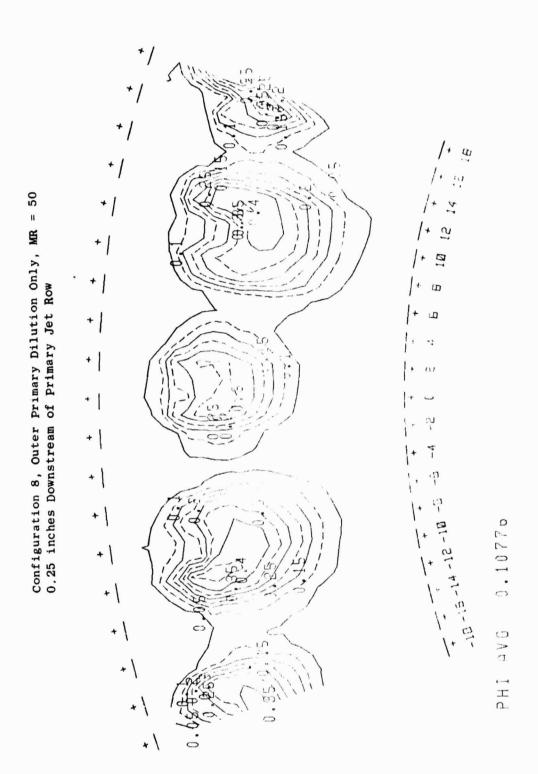
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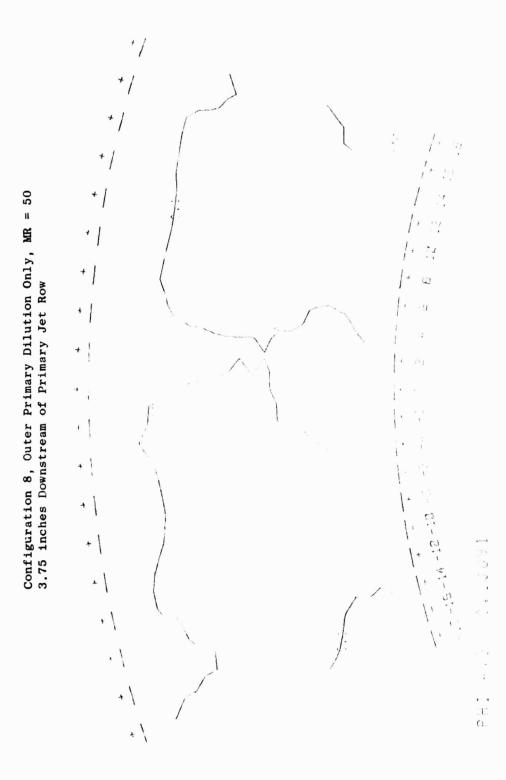
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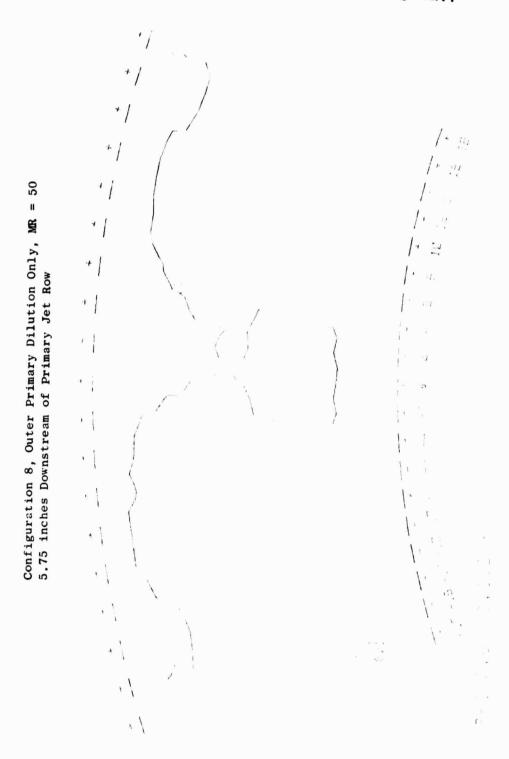


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18 Configuration 8, Outer Primary Dilution Only, MR = 50 1.75 inches Downstream of Primary Jet Row 5 V G Т. Г. ۸ \



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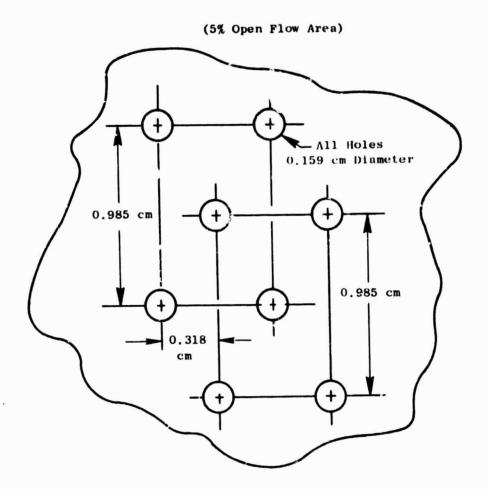
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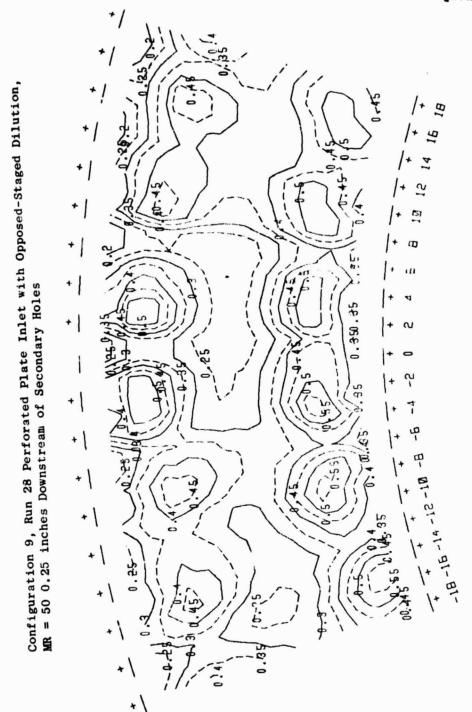
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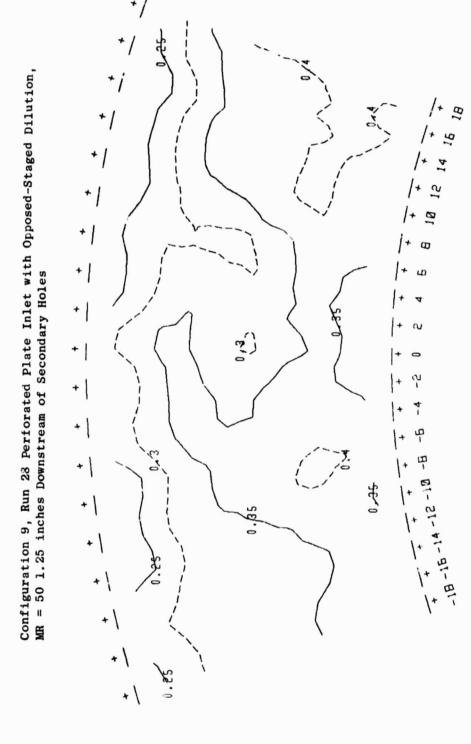
Typical Section of Perforated Plate



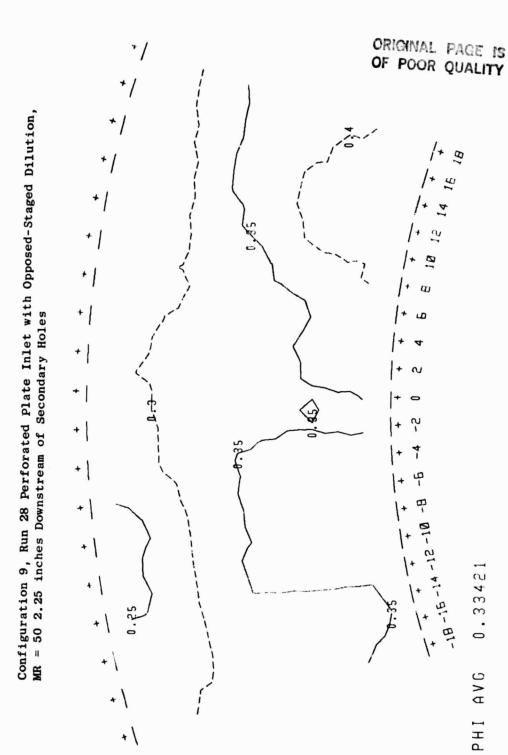
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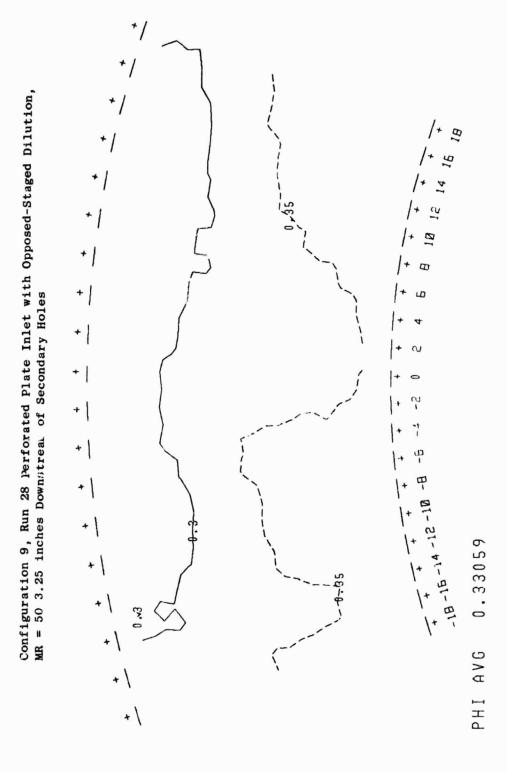


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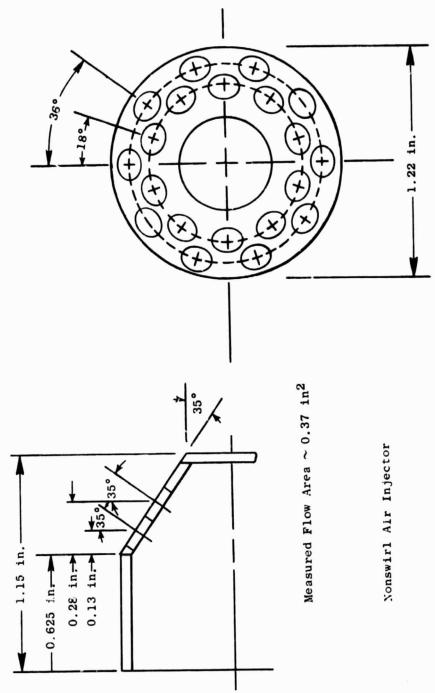


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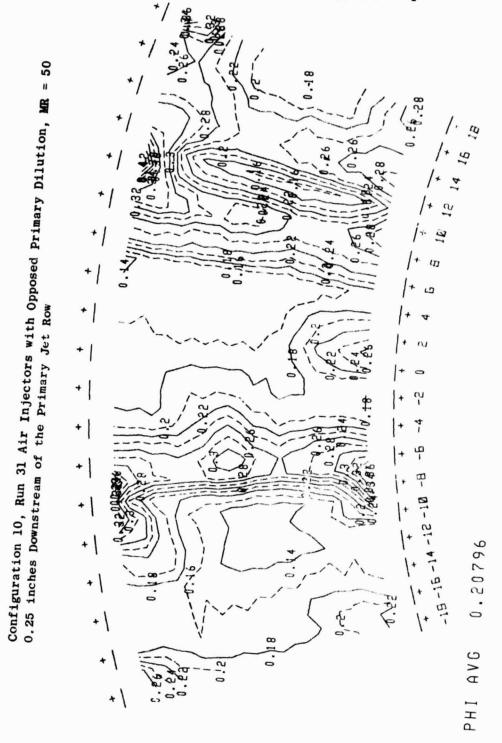
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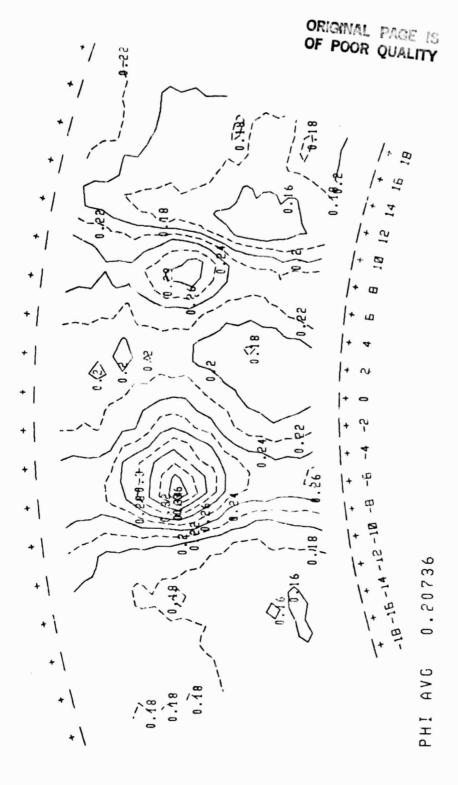
Each Row of Holes Contains 10 Equally Spaced Holes 3/16 in. Diameter Located As Shown



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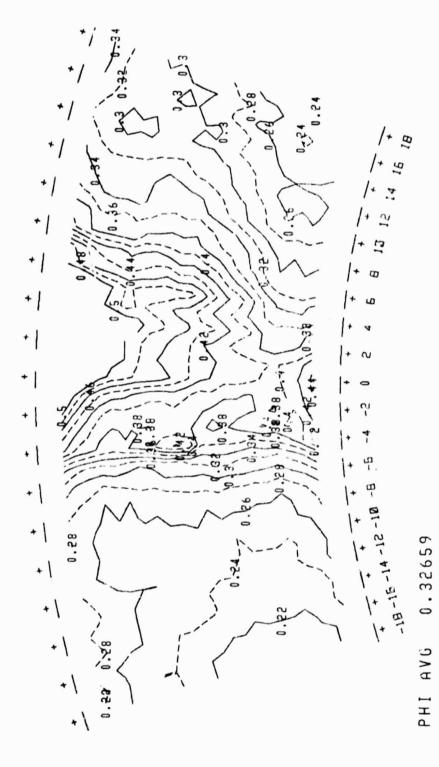
Configuration 10, Run 31 Air Injectors with Opposed Primary Dilution, MR = 50 1.75 inches Downstream of the Primary Jet Row



40.0 Configuration 10, Run 31 Air Injectors with Opposed Primary Dilution, MR = 50 3.75 inches Downstream of the Primary Jet Row 18 1,22.9 9 0.2592 AVG РНІ

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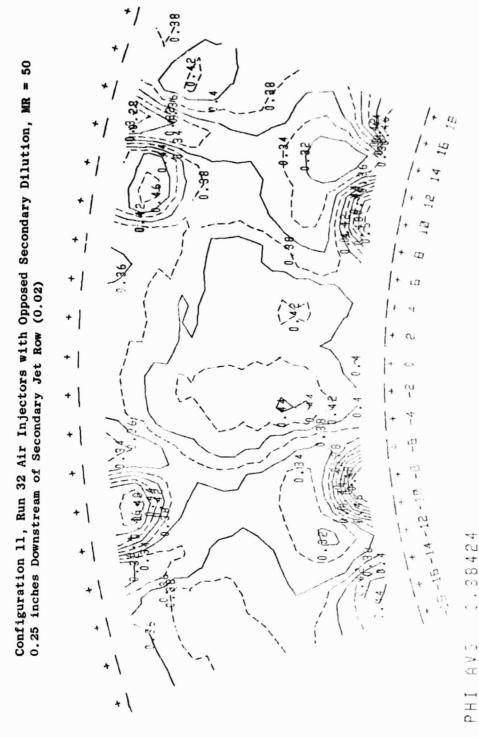


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EXPERIMENTAL CONFIGURATION 11	A DOME	STARTING R ER 3 DEG CW	SPACING EVERY 9 DEG STARTING AT - 5/16 INCH DIAMETER - 13/32 INCH DIAMETER	3 756 CW	Y = 19 FPS	1	VELOCITY =		GAS TEMPERAYURES	œ	TWIC4 NO	6	816. 815. 817. 816.	1	824. 620. 826. 821.	1	829. 326.		08/22/83	
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	JE 1	SPACING EVERY - 5/16 INCH D - 13/32 INCH - 18/56 STA	SPACING EV - 5/16 INC - 13/32 IN	EVERY 6 DEG STARTIN 23/64 inch diameter	DEG R,	:	585	5.5	MEASURE		SECONDARY	9	804. 811. 804. 812.	1.		١.	832. 855.	.		
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	OFPOSED N-SWIRL A	N: L: PRIMAR 0, 3, 36 D 9, 27, 45 U HOLE SPA R; ALL H	L: PRIMAR 0, 18, 36 D 9, 27, 45 D	ארר רני או	0.31 PPS AT	Σ		CHOMENTUM RATIO			ES DOWNSTREAM	6	5. 796. 3. 793.	١.		1.	6. 905.	. 1	NF 1.1	
	DESCRIPTION: OFPOSE FLOW FROM NON-SWIRL	HOLE PATTERN: CUTER WALL: PRIMARY HOLE HOLES AT 0, 13, 36 DEG HOLES AT 9, 27, 45 DEG SECONDARY HOLE SPACING TOP CENTER; ALL HOLES	INNER WALL: PRIMARY HOU HOLES AT 0, 18, 36 DEG . HOLES AT 9, 27, 45 DEG .	SECONDARY HO TOP CENTER; TEST CONDITIONS	CROSSFLOW: 0.31	5	INNER JET FLOW:	HOM)			0.25 INCHES	THE TA 2	1 797 795.	791.		808	8 773. 786		*** 51614EE/CONF11	

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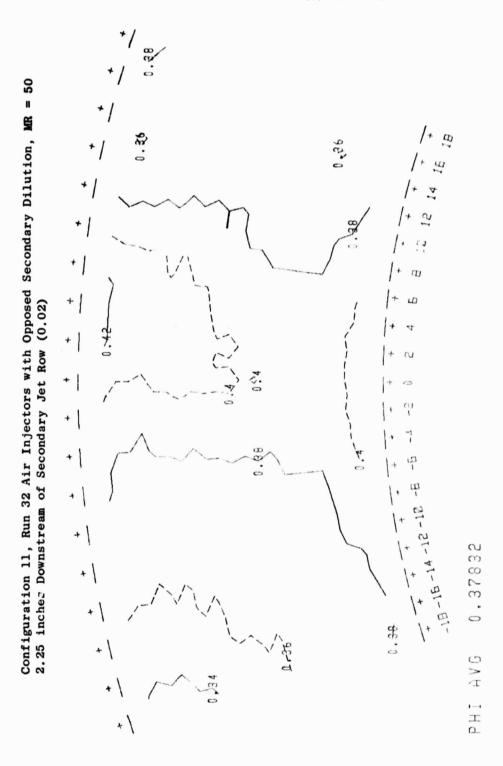
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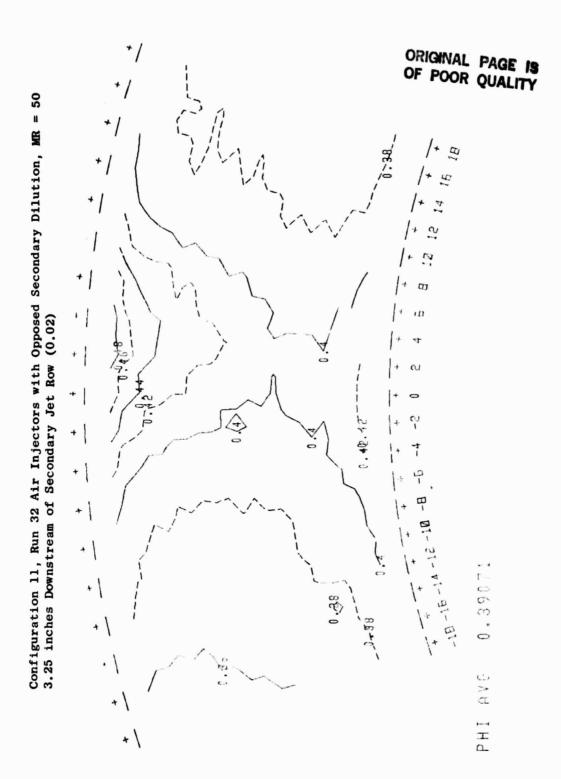


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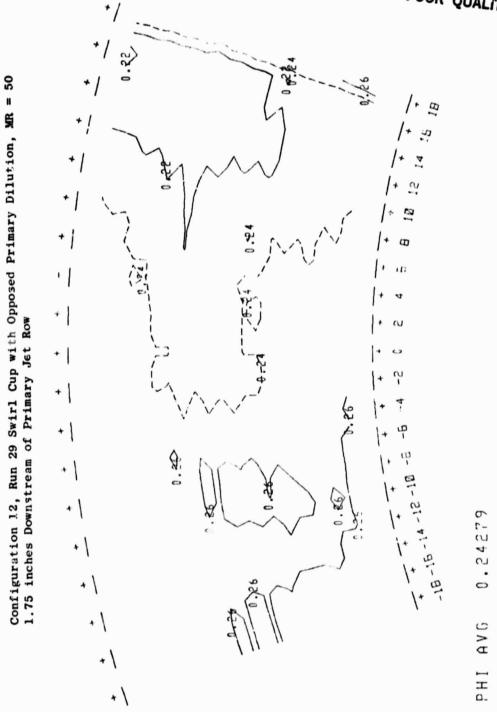
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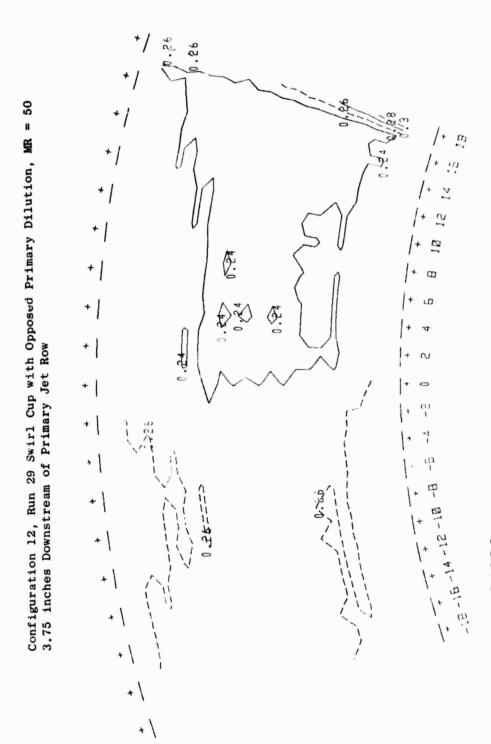
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Configuration 12, Run 29 Swirl Cup with Opposed Primary Dilution, MR = 50 0.25 inches Downstream of Primary Jet Row 0.-28 B m Ф 0 0.280.28 +\

PHI AVG 0.24879

ORIGINAL PAGE IS OF POOR QUALITY





PHI AVE 3.24978

0 x24 0.23 Configuration 12, Run 29 Swirl Cup with Opposed Primary Dilution, MR = 505.75 inches Downstream of Primary Jet Row B 7. 43.4 un 0.24986 ΑVG I H.d

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791.	.000	902	. 191	775	796.	805	789	783.	784.	792.	608	810	802	800	. 600	931	834	.618	796.	793.	803.			12	9	700		793	795.	797.	798.	797.	. 60		906	807	807	808	808	807	906	900					
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775.	.171	767.	780	783	775.	.69	769.	769.	782.	784	776.	770.	767		785	786.	786.	795.	803	810	800.		OF SECO	10	701		784	785.	786.	786.	787.	788.	788	200	789.	290	789.	791	. 167	792.	793	1 80		795			
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904	SECONDARY	9	785.	785.	786.	786.	787.	787.	788.	789.	789	700	792.	791	792.	793.	793.	793.	794	793.	784	795	794	794	795.	794	794.	794.	7 0 2	794	795.	795.	796.	795	794.					
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799.	DCWNSTREAM	6	780	779.	780.	779.	780	781	780.	782	781	782	782	780.	783.	784.	782.	784	784	782	7 03	784	784	785.	785.	783.	785.	787.	788	788	789.	790	791	791	790		6			
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Configuration 13, Run 30 Swirl Cup with Opposed-Staged Dilution, MR = 50 0.25 inches Downstream of Secondary Jet Row B _{√} 00 0.4 ΑVG H H n.

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1.36 98.0 Configuration 13, Run 30 Swirl Cup with Opposed-Staged Dilution, MR = 50 1.25 inches Downstream of Secondary Jet Row 0.387=5 ΑVG РНІ

Configuration 13, Run 30 Swirl Cup with Opposed-Staged Dilution, MR = 50 2.25 inches Downstream of Secondary Jet Row



Configuration 13, Run 30 Swirl Cup with Opposed-Staged Dilution, MR = 50 3.25 inches Downstream of Secondary Jet Row 18 0 ノノノノノノルサンタノーノノノ + 9 + 0.39616 9 N G H. D. ٠\

DESCRIPTION: OPPOSED AND STAGED JET ROWS FLESSIRES EQUAL HOLES AT 0.18, 38 DEG 57,16 INCH IN HOLES AT 0.18, 38 DEG 13/32 INCH IN HOLES AT 0.27,45 DEG 13/32 INCH IN HOLES AT 0.27,45 DEG 13/32 INCH IN HOLES AT 0.27,45 DEG 13/32 INCH IN HOLES AT 0.18, 38 DEG 13/32 INCH IN HOLES AT 0.18, 38 DEG 13/32 INCH IN HOLES AT 0.18, 38 DEG 13/32 INCH IN HOLES AT 0.27,45 DEG 13/32 INCH IN HOLES AT 0.27,45 DEG 13/32 INCH IN HOLES AT 0.27,45 DEG 13/32 INCH IN HOLES AT 0.27,45 DEG 13/32 INCH IN HOLES AT 0.11 PPS AT S50 DEG R, VELO OUTER JET FLOW: 0.11 PPS AT S50 DEG R, VELO OUTER JET FLOW: 0.11 PPS AT S50 DEG R, VELO OUTER JET FLOW: 0.11 PPS AT S50 DEG R, VELO OUTER JET FLOW: 0.13 PPS AT S54 DEG R (MOMENTUM RATIO - 766:1) INNER JET FLOW: 0.13 PPS AT S54 DEG R (MOMENTUM RATIO - 77:1) MEASURED O 25 INCHES DOWNSTREAM OF SECONDARY IN ETA 1 2 3 4 5 6 O 665. 664. 661. 660. 658. 658. 65 1 663. 661. 650. 654. 651. 642. 642. 643. 643. 643. 643. 643. 643. 643. 643

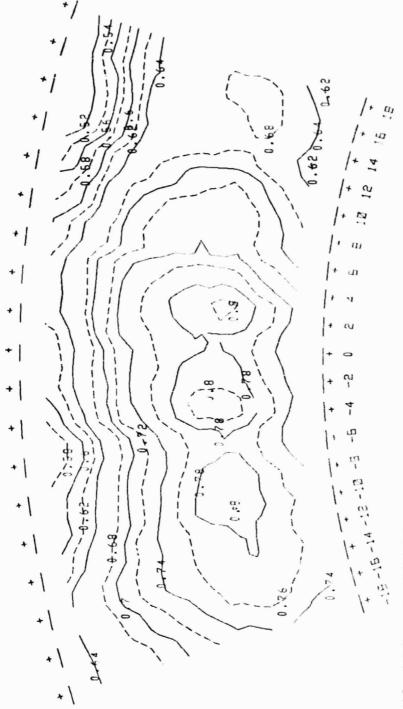
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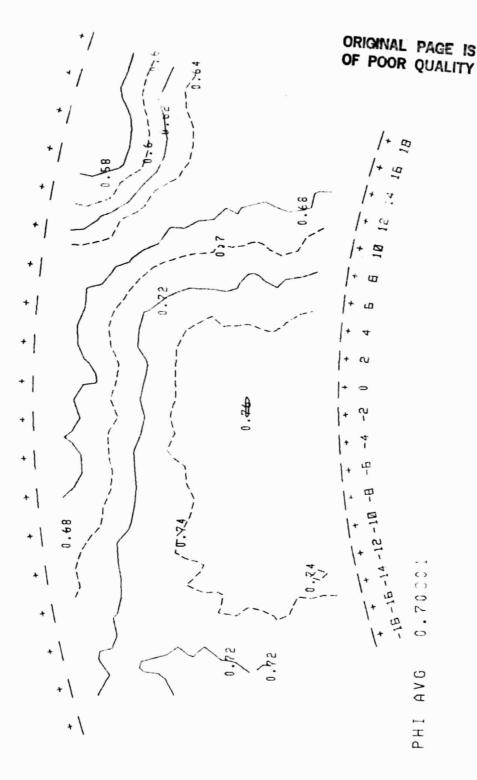


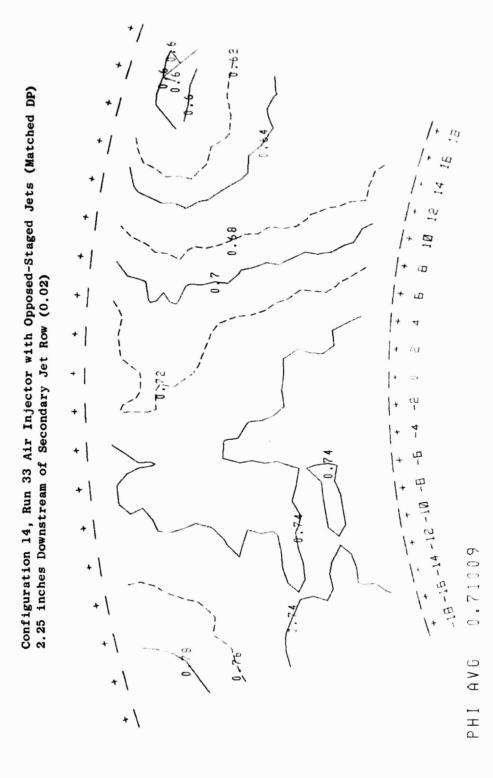
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Configuration 14, Run 33 Air Injector with Opposed-Staged Jets (Matched DP)

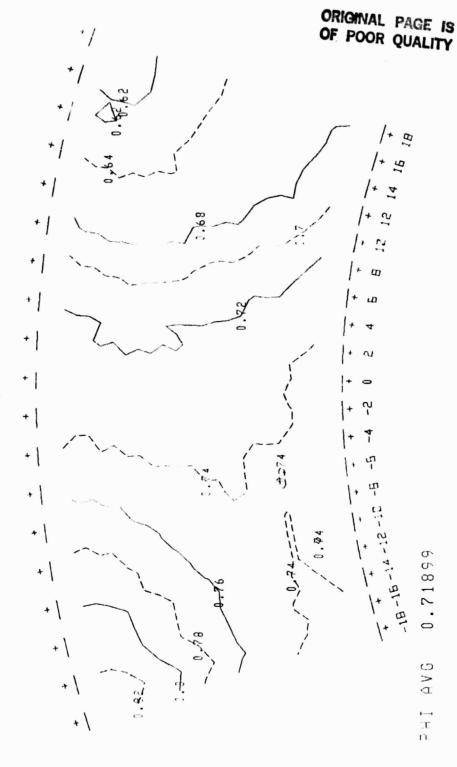
0.25 inches Downstream of Secondary Jet Row (0.02)

Configuration 14, Run 33 Air Injector with Opposed-Staged Jets (Matched DP) 1.25 inches Downstream of Secondary Jet Row (0.02)



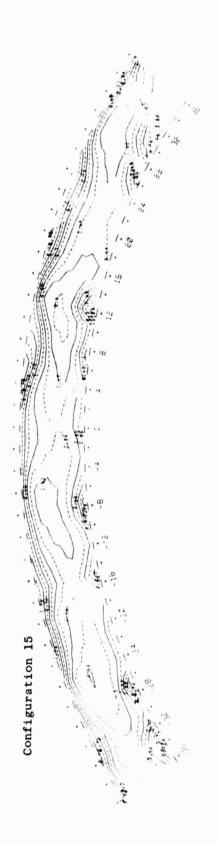


Configuration 14, Run 33 Air Injector with Opposed-Staged Jets (Matched DP) 3.25 inches Downstream of Secondary Jet Row (0.02)



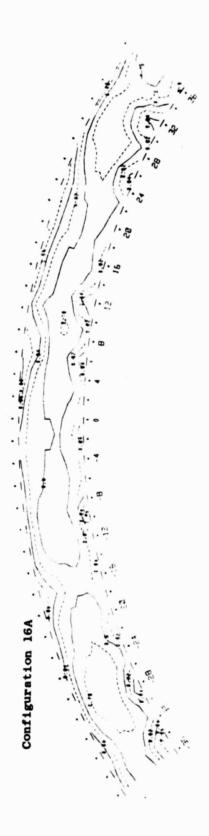
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EXPERIMENTAL CONFIGURATION 15 STOR TWO TEMPERATURE ON			DEG R	N VELOCITY 168 FPS	N VELOCITY 172	ES R	O,				22/83	
EXPERIMENSECTOR COMBUSTOR TWO	FINITIONS .		PPS; 958	EG R	× 7	DEGRE	6 7		696. 696. 701. 700. 707. 707.			
EXI 90 DEG SECTÖR COMBUSTO STANDARD CONFIGURATION	HOLE SIZE DE		; FLOW 0.365 DITIONS:	PS AT 537 DEG	1		4	- 10	686. 695. 697. 701. 698. 705.	l		
F101	L L	ST CONDITIONS:	DELTA P/F 4.21 %; FLOW	AL FLOW 0.69 PPS AT	INNER LINER DELTA		, n	60 60	674. 661. 674. 684. 686. 693. 678. 689.		/CONF15	
DESCRIPTION:	HOLE PATTERN: SEE ATTACHME MEASURED FLO	TEST CONDI	DELTA P/P	TOTAL F	INNE		THETA	662.	2 637. 3 669. 4 679. 5 658.		5	

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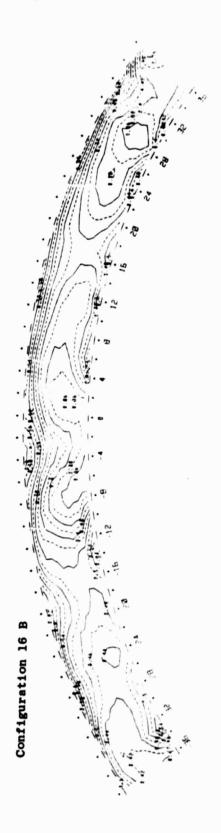
SCRIPTION: F101 90 DEG SECTOR CACE EXPERIMENT. STANDARD CONFIGURE OLING. DOME AND LINER PRESSURE TO CALING. SEE ATTACHMENT FOR: MEASURED FLOW AREAS/DISTRIBUTION	E/CONFIGA 08/22/83 8.144 PAGE 1 ***	08/22/83 8.144 PAGE 1 **	621. 625. 628. 629. 627. 627. 623. 621. 617. 608.	621. 627. 630. 631. 629. 628. 625. 622. 619.	627. 629. 632. 633. 634. 632. 633. 630. 629. 626. 627. 603.	630 630 635 636 635 635 637 63 638	614, 624, 629, 634, 634, 636, 633, 632, 629, 621, 609,	621. 625. 629. 632. 633. 633. 630. 627. 624. 614. 604.	623. 627. 629. 631. 631. 630. 630. 626. 627. 614. 604. 534.	396. 600. 608. 619. 623. 626. 627. 624. 623. 620. 611. 600. 608. 615. 619. 626. 627. 628. 624. 621. 621. 622. 627.	602. 605. 616. 622. 626. 626. 627. 624. 622. 620. 612. 601.		3 4 5 6 7 6 9 10 11 12 13	OR	DEGREES R	MEASURED GAS TEMPERATURES	NNER LINER DELTA P/P 1.13 %; INJECTION VELOCITY 145 FPS	JTER LINER DELTA P/P 1.10 %; INJECTION VELOCITY 143 FPS	COMBUSTOR LINER CONDITIONS	INLET CONDITIONS:		RED FLOW AREAS/DISTRIBUTION	ATTERN: TTACHMENT FOR:	LINER PRESSURE DROPS EAUA	TION: FIOI 90 DEG SECTOR COMBUSTOR TWO TEMPERATURE XPERIMENT: STANDARD CONFIGURATION, WITHOUT SPLASHPLATE	EXPERIMENTAL CONFIGURATION 15A
DESCRIPTION: F101 TRACE EXPERIMENT. COOLING. UOME AND HOLE PATTERN: SEE ATTACHMENT FO MEASURED FLOW ARE DELTA P/P 1.11 COMBUSTOR LINER OF TOTAL FLOW 0.57 OUTER LINER DE 1 590. 596. 605. 1 590. 596. 605. 2 603. 604. 615. 5 607. 614. 621. 6 607. 614. 621. 10 609. 617. 621. 6 607. 614. 621. 10 609. 617. 621.	51814EE/CONF16	51814EE/CGNF16A	608. 614.	610. 617.	621. 627.	621. 625.	607. 614.	614. 621. Ro7 606	619. 623.	603. 608.	592, 602.	:	-				INNER LINE	OUTER LINE	COMBUSTOR LINE	DOME INLET CO		MEASURED FLOW	HOLE PATTERN: SEE ATTACHMEN'	COLING. COME	ESCRIPTION: F	

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MENTAL CONFIGURATION 168	TWO TEMPERATURE WITHOUT SPLASHPLATE DRAPS							FLOCITY 96 FPS	96	1PERATURES	œ	01 6	5. 783. 778. 7. 784. 783.	781.	769.	760.	762.	752.	08/22/83 8.144	
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	F101 90 DEG ENT. STANDAR ATCHED DOME A	ENT FOR HOLE		NS:	9 98 T. FI ON		0.36 PPS AT	OUTER LINER DELTA P/P	INNER L!NER DELTA P			6	~ ~ !	765.		775	778.	Ы		
	DESCRIPTION: F10 TRACE EXPERIMENT COOLING, UNMATCH	HOLE PATTERN: SEE ATTACHMENT FOR HOLE SIZE DEFINITIONS PIEACHED FLOW AREAS/DISTRIBUTION		TEST CONDITIONS:	DEI TA P/P	COMBUSTOR LI	TOTAL FLOW	OUTER L	INNER L			Theta i 2	787. 792	745.	746.		772.	.	*** 51814EE/CONF168	

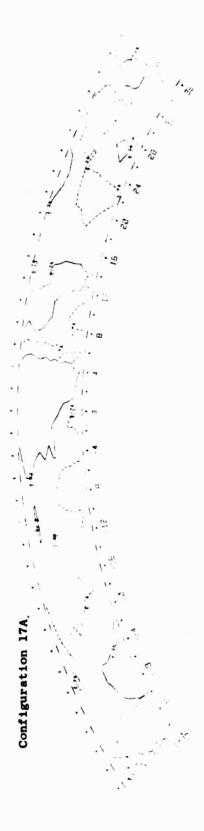
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ING TEMPERATURE WITHOUT DOME OR LINER					ON VELOCITY 144 FPS	ON VELOCITY 145 FPS	MEASURED GAS TEMPERATURES		۰	653	644	639	643	644.	644	647	648.	2/83	
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DESCRIPTION: FIOI 90 DEG SECTOR COMBUSTOR TWO TEMPERATURE TRASE EXPERIMENT. STANDARD CONFIGURATION. WITHOUT DOME OR LINER COOLING. EQUAL DOME AND LINER PRESSURE FROPS.	HOLE PATTERN: SEE ATTACHMENT FOR HOLE SIZE DEF MEASURED FLOW AREAS/DISTRIBUTION	TEST CONDITIONS:	DELTA P/P 1.	TOTAL FLOW	T U O	Ž			+	6.14	638. 637.	637.	636	635	639	633	9	51814EE/CONF17A	
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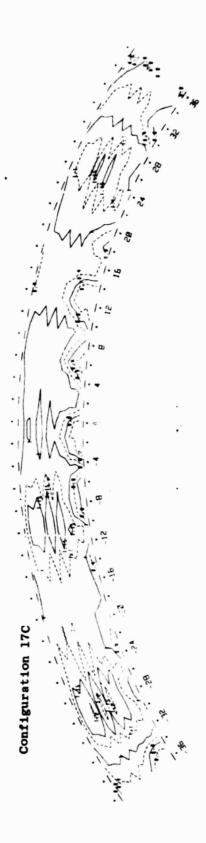
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ENTAL CONFIGURATION 178	TWO TEMPERATURE WITHOUT DOME OR LINER DROPS.						96 FPS	6 FPS	ES		0	811.	8 8 8 8	814	821	824	823	818	8.14	
CONFI	IPERATU JT DOME						VELOCITY 9	VELOCITY 96	TEMPERATURES		o	912.	816.	815	821.	822	822	818	/22/83	
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	G SECTOR ARD CONF	E SIZE	STRIBU			AT 535	P/P 0.	P/P 0.			ID.		815.		821.		815			
	DESCRIPTION: F101 90 DEG SECTOR COMBUSTOR TRACE EXPERIMENT. STANDARD CONFIGURATION. COOLING ON LINER PRESSURE	OLE PATTERN: SEE ATTACHMENT FOR HOLE SIZE DEF	EASZEI		ĸ	0.18 PPS					4		7. 816.	١.	3. 822.					
	F101 MENT.	IN:	3V MO":	 ONS	<u> </u>	0	LINER DELTA	INNER LINER DELTA					817. 817. 807. 806.			800		9 823 5. 827	INF: 78	
	DESCRIPTION: TRACE EXPERI	PATTER	URED F	CONDITIONS	DELTA P/P		OUTER	INNER			8			ı		792 796		825	51814EE/CONF:78	
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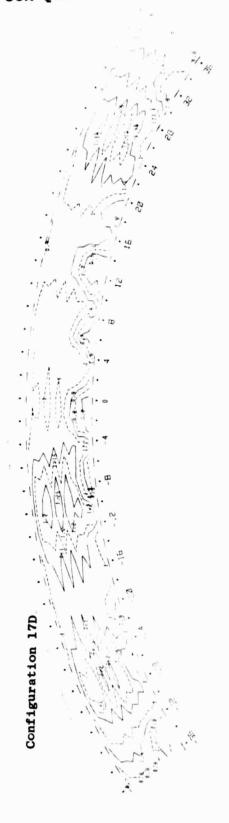
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EXPERIPION: F101 90 DEG SECTOR COMBUSTOR. STANDARD CONFIGURATION W/O DOME & LINER COOLING FUEL INJECTED USING FTOT HIGH PRESSURE ATOMIZING FUEL NOZZLES HOLE PATTERN: SEE ATTACHMENT FOR: NEA:URED FLOW AREAS/DISTRIBUTION	DELTA P/P 5.80 % COMBUSTOR LINER CONDITIONS TOTAL FLOW 0.64 PPS AT 1360 DEG R OUTER LINER DELTA P/P 5.76 Z; INJECTION VELOCITY 526 FPS INNER LINER DELTA P/P 5.75 X; INJECTION VELOCITY 526 FPS EXIT FUEL-AIR RATIO 0.016	MEASURED GAS TEMPERATURES DEGREES R	THETA 1 2 3 4 5 6 7 6 9 10 11 12 13 0 2045. 2036. 2032. 2090. 2213 2293. 2299. 2354. 2303. 2336. 2313. 2303. 2209. 1 2169. 2148. 2165. 2256. 2325. 2361. 2366. 2413. 2369. 2370. 2370. 2418. 2169. 2466. 2475. 2256. 2361. 2461. 2505. 2481. 2471. 2353. 3 2518. 2461. 2505. 2481. 2471. 2353. 3 2518. 2461. 2508. 2569. 2569. 2569. 2561. 2570. 2612. 2569. 2569. 2569. 2569. 2614. 2565. 2569. 2659. 2657. 2614. 2657. 2617. 2617. 2617. 2617. 2617. 2617. 2618. 2618. 2619

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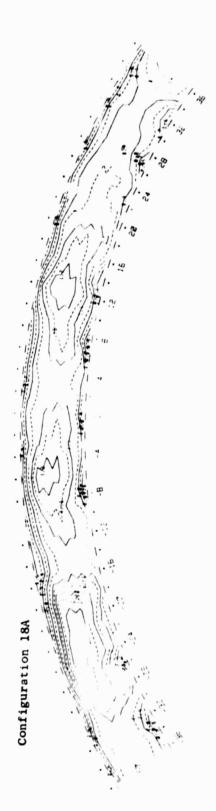
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DESCRIPTION: F101 90 DEG SECTOR CI CONFIGURATION. FUEL INJECTED USIN FUEL NOZZLES HOLE PATTERN: SEE ATTACHMENT FOR: MEASURED FLOW AREAS/DISTRIBUTION	TEST CONDITIONS: DELTA P/P 11.90 % COMBUSTOR LINER CONDITIONS	TOTAL FLOW 0.64 OUTER LINER DE INNER LINER DE			1 2064. 2013. 1 2062. 2013. 2 2163. 2137. 3 2449. 2477. 4 2559. 2610. 5 5516.	2410. 2410. 2411. 2448. 2320.	

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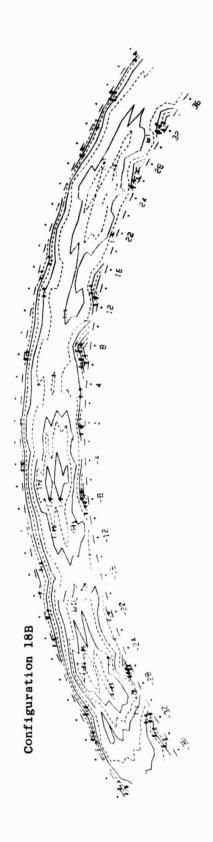
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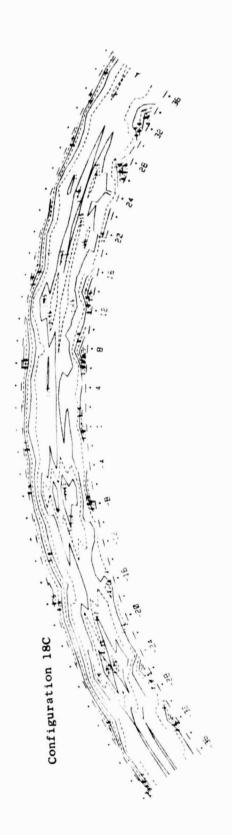
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DESCRIPTION: F101 90 DEG SECTOR	HOLE PATTERN: SEE ATTACHMENT FOR: MEACURED FLOW AREAS/DISTRIBUTION	TEST CONDITIONS: DOME INLET CONDITIONS	DELTA P/P 5 %	TOTAL AIRFLOW 0.98 PPS	INLET TEMPERATURE 1360 DEG R	CITY GAS			THETA 1 2 3 4 5	0 2059, 2179, 2219, 2264, 2287, 1970, 2226, 2300, 2355, 2373, 2 1923, 2145, 2283, 2401, 2429	2204. 2314. 2412. 2475. 2482. 2543. 2563. 2604. 2591. 2675. 2662. 2711.	2073. 2375. 2495. 2631. 2420. 2546. 2614. 2683. 2372. 2533. 2553. 2644.		

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EXPERIMENTAL CONFIGURATION 18 DESCRIPTION: F101 90 DEG SECTOR COMDUSTOR. STANDARD CONFIGURATION FUEL 1 JECTED USING F101 HIGH PRESSURE ATOMIZING MOZZLES	HOLE PATTERN: SEE ATTACHMENT FOR: MEASTIRED FLOW AREAS/DISTRIBUTION	TEST CONDITIONS:	DELTA P/P 5 %	3	JP-5 FUEL MEASURED GAS TEMPERATURES	DEGREES R	1 2 3 4 5 6 7 9 9		2443. 2538. 2647. 2628. 2702. 2633. 2712. 2539. 2552. 2625. 2629. 2693. 2627. 2696. 2672. 2669. 2736. 2719. 2772. 2685. 2726.	2397, 2597, 2633, 2739, 2745, 2817, 2745, 2804, 2694, 2120, 2403, 2601, 2755, 2760, 2835, 2770, 2841, 2737, 2865, 2785	2739. 2744. 2627. 2806. 2869. 2759. 2834.	EE/CONF18C 08/22

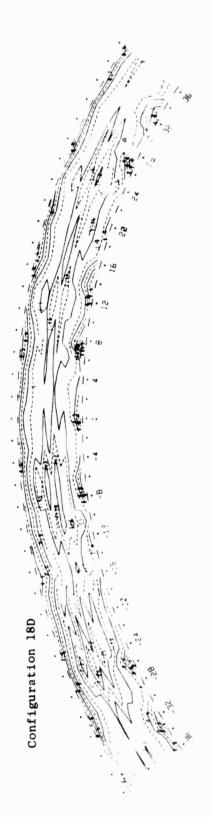
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